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US Department of Transportation	n	(Airfran	ne, P	owerplant, Pr	op	ell	er, or App	iance)	-		For FAA Use Only
Federal Aviation		•	•	,			· · · · · ·	- 7			
INSTRUCT	IONS: Print	or type al	l entri	es See Title 14 (CEF	3 8	43.9 Part 43	Appendix B and		3.9-1 (or sub	sequent revision thereof) for
instructions	and dispos	ition of this	form.	This report is req	uire	ed b	by law (49 U.S	S.C. §44701). Fai	lure to	report can re	sult in a civil penalty for each
such violati	on. (49 U.S.			- Maarla							
	USA	y and Regi	stratio	п магк N5216	6E			Serial No. 24-1658			
1. Aircraft	Make							Model		Ţ:	Series
	Moone	ey						M20J			
	Name (As	s shown on	regist	ration certificate)	_			Address (As sl	hown oi	n registration	certificate)
2. Owner	DOEF	RING RU	SSE	LF				Address 751 Sil		ek Dr	
								city Leande	er.		_{State} _TX try_USA
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						<u>3. r</u>	For FAA Use				<u> </u>
4. Tyj	oe	[5. I	Unit Identifica	tion			
Repair	Alteration	Uni			Mał	ke			Model		Serial No.
	X	AIRFRAM	E			_		(As described	a in Iten	n 1 above) 	
		POWERP	LANT								
		PROPELI	ER								
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			-	l <u> </u>	6	Co	onformity Stat	ement	_		
A. Agency's N	ame and A	ddress			<u> </u>		Kind of Agend				
Name St. Cl	oud Aviation	n			_		U.S. Certific	ated Mechanic		Mar	ufacturer
Address 1544		e SE		State MN	-			ficated Mechanic		C. Certif	icate No.
city <u>Saint</u> zip 56304		untry USA		_ State <u>MN</u>	-	X		Repair Station		BDKR3	93C
								laintenance Organi			
have b	een made in	accordanc	e with	tion made to the u the requirements to the best of my ki	of F	Parl	t 43 of the U.S				erse or attachments hereto at the information
Extended ran				ature/Date of Author			<u> </u>				<u></u>
per 14 CFR F								-			
Арр. В			Auç	gust 10, 2017			James D Fa	airchild 🖌	~>>	OF	instart
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Certificate or Designation N			Signa	ature/Date of Auth	oriz	ed	Individual	_		<u> </u>	· / · //
BDKR3	893C		Aug	gust 10, 2017			James D Fa	airchild <u>h</u>	\sim	DIA	minin
FAA Form	337 (10-06)										

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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.
8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)
USA N5216E 08/10/2017
Nationality and Registration Mark Date
 Removed the following: A. Garmin GTX330 Transponder B. Terra AT3000 Encoder C. II Morrow 428-2003 Encoder
 2. Installed the following: A. Garmin GTX345 ADS-B transponder B. ACK A-30.9 Altitude Encoder
3. Installed the GTX345 in radio stack using provisions from the removed GTX330. Replaced existing coaxial cable and wired from existing antenna to the GTX345. The GTX345 gets GPS position information from a previously installed Garmin GNS530W. The GTX345 provides FIS-B weather and TIS-B traffic to the GNS530W and also through Bluetooth. The GTX345 gets altitude info from new A-30.9 digital encoder w/ RS-232 found under the panel. The GTX345 is wired to the avionics buss through a 7277-2 series Klixon breaker. The A-30.9 is mounted at the pilots side firewall in the same location as the removed 428-2003.
4. The GTX345 was installed IAW Garmin install manual p/n 190-00734-10 Rev. 8 and per STC SA01714WI. The A-30.9 was installed in accordance with ACK Technologies install manual p/n A30M Rev. 6 and meets FAA TSO C88a. All units were installed per AC43.13-1B chapters 4,7,10,11 and 12 and AC43.13-2B chapters 1,2,3 and 11.
 Installed Garmin FAA approved Flight Manual Supplement GTX345 p/n 190-00734-15 Rev. 2 in the aircraft flight manual.
 Included Garmin GTX345 instructions for continued airworthiness p/n 190-00734-11 Rev. 5 section 4, in the aircrafts maintenance records.
Performed Transponder, altimeter and encoder test as required by FARs 91.411, 91.413 and 91.217 and were found to be within the limits of FAR43 appendices E and F and AC43.6b.
8. As installed the GTX345 ADS-B OUT system meets the requirements of 14 CFR section 91.227.
9. The continuous electrical load does not exceed 80% of the charging system.
10. Updated the weight and balance and equipment list.
ENDEND
Additional Sheets Are Attached

FAA Form 337 (10-06)

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Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED FAA Fit. Standards -Manufacturer -X -Maintenance Organization Person Appoved by Canadian Department of Transport Y FAA Designee X Repair Station Inspection Other (Specify) ertificate or Designation No. Signature/Date of Authorized Individual Dave Backes 12/28/2007						· · · · · · · · · · · · · · · · · · ·		Dav	/e Backe	s 12/	28/2007
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FAA Fit. Standards Inspector Manufacturer -X Maintenance Organization Person Appoved by Canadian Department of Transport SY FAA Designee X Repair Station Inspection Authorization Other (Specify) iertificate or Designation No. Signature/Date of Authorized Individual Dave Backes 12/28/2007	Pursuar the Adm	nt to the au	ithority given p of the Federal	ersons specified below Aviation Administration	, the and	e unit identifi d is 🔽 APPF	ed in item ROVED	5 was ins			nner prescribed by
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DKR393C Dave Backes 12/28/2007	- F	•						tion I	• • •		·····
	-			Signature/Date of AUINC	JIIZE			Backes	12/20/20	007	• •
		(10-06)					Dave	DAUKES	12/20/2	007	

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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

12/28/2007

Nationality and Registration Mark

Date

1. Description:

Upgraded the GNS530 to a GNS530W according to STC SA01933LA. Installed a GA35 GPS antenna on top of the cabin. Reused the existing nav and com antenna systems. Enclosed is an FAA Approved Flight Manual Supplement Dated . Certified the GNS530W for IFR enroute and for precision approaches.

Weight and balance negligible. The equipment list has been revised. The change in the electrical load does not exceed 80% of the charging system. The airplane has been test flown according to FAR 91.407 and is returned to service.

3. Instructions of Continued Airworthiness:

Included Garmin document p/n 190-00357-65 rev A for the GNS530W Instructions for Continued Airworthiness in the aircraft maintenance records. This supersedes ICA for the previously installed GNS530.

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1. Aircraft	Make Mooney				Model M20				-	
	Serial No. 24-1658				Nation N52		egistration M	ark		
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hereto have information	the repair and/or alteration made to been made in accordance with the r furnished herein is true and correct to	equirements o the best	nts of r	of Part 43 of th	he U.S.	Federal	Aviation R	the rev egulatio	verse or attac ons and that t	hments he
12	-30-03			or Return To	\wedge	lour	Barby		Dav	ve Backes
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BY FAA Inspe	Flt: Standards Manufacturer			Inspection A Person Appove	uthoriz		Other (S			
Date of Approva				Canada Airwon Signature	thiness (Group	 dividual			
AA Form 337		-	1	· · · ·						

NOTICE Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements. 8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

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1. Introd	uction			
		r 11 Chapter AC43 13-2A	chapter 1 par.1 - thru 11, Chapt	ter 2 nar 21 thni 27
Manufact	=		Install Man P/N	tor 2 put. 21 tinu 27.
Garmin	GTX330	Transponder Mode S	190-00207-02 rev E	
Quinni	0111000	Thisponder Mode o		
2. Descri	ption: Installed the	transponder in the radio stat	ck. Removed a KT76A transpon	der. Reused the existing antenna
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.3. Contro	l, operation inform	ation. Conjult the oppropri	into monitori foloritario o monitori uno	mual.
4. Servici	ing information: N/	A 2 70 20365	TTE ET EL COMPLETE	
5. Mainte	enance instructions	N/A		
6. Troub	le shooting informa	ition: N/A	1.46	
7. Remov	al and replacemen	t information: Consult the	e appropriate manufacture's insta	all manual.
8. Diagra	ms: N/A			
9. Specia	l inspection require	ements: N/A		
	ation of protective	treatments: N/A		
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Date of	f Approval	or Rejection	•	Certificate or De	signation N	l o.		Signature			vidual			
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FAA Form 337 (12-88)

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NOTICE
Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.
8. Description of Work Accomplished
(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)
N5216E M20J SN 24-1658
Installed STC SA5631SW Standby vacuum system in accordance with Aero Safe drawing list 820900 revision C dated 4-19-85 or later FAA approved revison. Weight and Balance amened
END

DIECIEDVIED SEP 3 0 2002 MSP FSD0
Additional Sheets Are Attatched
Reproduced by Avantext, Inc. *U.S.GP0:1994-568-012/0001

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Administration							N	<u>ISP I</u>	FSDO	
instructions and	disposition of this fo	ntries. See FAR 43.9, F. orm. This report is requir on (Section 901 Federal	ed by	/ law (49 U.S.	C. 142					
1. Aircraft	Make Mooney				Mode M20					
	Serial No. 24-1658	<u></u>				nality and Re 216E	gistration Ma	rk		
		n registration certificate)				ess (As show	n on registrati	on certi	icate)	
2. Owner	Russell Doeri				365	2 County arlake MN	Rd 8		,	
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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

1. Introduction:

Reference AC43.13-1B Chapter 11, Chapter, AC43.13-2A chapter 1 par.1 - thru 11, Chapter 2 par. 21 thru 27, AC20-138 and used STC SA00864WI for basis of the installation. The GPS has TSO-C129a Class A1.

 Equipment
 Description
 Install Man P/N

 GNS530
 GPS/NAV/COM
 190-00181-02 rev B

2. Description: Removed a KX155 nav/com and a GPS150XL. Installed a Garmin GNS530 GPS/NAV/COM in the radio stack. The GPS is wired to the HSI and the Century 2000 autopilot. The GPS antenna is mounted on top of the cabin. Enclosed is an FAA Flight Manual Supplement Dated $\frac{2}{2}$. The system is certified for IFR enroute and Non-precision approaches.

Weight and balance and the equipment list has been revised. The change in the electrical load does not exceed 80% of the charging system.

(I certify the plane has been test flown and meets the requirements of AC20-138 and FAR 91.407.

8-6-02 Date Certificate Number Pilot's Name

- 3. Control, operation information: Consult the appropriate manufacture's install/operators manual.
- 4. Servicing information: N/A
- 5. Maintenance instructions: N/A
- 6. Trouble shooting information: N/A
- 7. Removal and replacement information: Consult the appropriate manufacture's install manual.
- 8. Diagrams: N/A
- 9. Special inspection requirements: N/A
- 10. Application of protective treatments: N/A
- 11. Data: N/A
- 12. List of special tools: N/A
- 13. For commuter category aircraft: N/A
- 14. Recommended overhaul periods: N/A
- 15. Airworthiness limitation section: N/A

16. Revision: A letter will be submitted to the local FSDO with a copy of the revised FAA form 337 and revised ICA. The FAA inspector accepts the change by signing Block 3 and including the following statement:

"The attached revised/new Instructions for Continued Airworthiness dated

for the above aircraft of component major alteration have been accepted by the FAA, superseding the Instructions for continued Airworthiness dated ______. Once the revision has been accepted, a maintenance record entry will be made, identifying the revision, it's location, date of the Form 337."

Additional Sheets Are Attatched

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2										I	Form Ap	proved	
	Department			MAJOR REF						4	JWR NO	.2120-0020 For FAA Use O	niv.
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	al Aviation											FSDO-07 ita, KS	IR9
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and o	aispositio	n of this form. This violation (Section 9	s rej	oort is required by	/ law (49	U.S.	6.C. 1	421). Failure	to report can re	sult in a civ	/il penalt	y not to excee	d \$1,000
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BY FAA Fit Standards Manufacturer X Inspection Authorization Other (Specify)													
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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

N5216E STC-SA00071SE 10-16-2000 INSTALLED COCKPIT SUN VISOR IN ACCORDANCE WITH FAA-APPROVED ROSEN DRAWING LIST NUMBER RMY-00DL, REVISION A, DATED OCTOBER 4, 1993.

Additional Sheets Are Attached

US Departmen of Transportation Federal Aviat Administration	ion (Airf	MAJOR REP rame, Powerp	AIR AND ALT plant, Propelle	FERATION r, or Applia	nce)		oproved lo. 2120-0020 For FAA Use Jentification	
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	Serial No. 24-10	558		Natio	onality and Regis	Iration Mark		
2. Owner	Name (As show	n on registration c	ertilicate)	Addr	N5216E ess (As shown or	registration	Certificate	
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			3. For F/	AA Use Only	HITA KS 6	7206-401	2	
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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

The following Century 2000 Autopilot serial #2K-0059 was installed in this Aircraft using Install Kit AK967-2030 with no Form 337. All components of this System is shown on Weight And Balance dated 14 SEPT 1988.

Performed conformity check of this Autopilot using Century Bulletin #2040 and per STC SA34462SW-D. The Autopilot is connected to the Aircraft electrical buss thru a 5 amp circuit breaker. The Electric Trim System is connected to the Aircraft electrical buss thru a 3 amp circuit breaker switch. The HSI is connected to the Aircraft electrical buss thru a 3 amp circuit breaker.

TABLE OF COMPONETS

PART #	<u>ITEM</u>	SERIAL #	LOCATION INS.AFT OF DATUM
NSD360A	HSI · ·	Did not remove.	11.25
IC948-2	NAV Switching Relay	56	.50
1B495	Flux Detector	H05295A	161.0
1D755	Slaving Adapter	912BR	117.85
23-1100-4A	26VAC Converter	None	117.25
1D937	A/P Computor	119	8.90
1B759-2	Disconnect Relay	233	6.25
10784-880	Roll Servo	541	59.75
10784-2-880	Pitch Servo	349	143.50
10791-3-884	Trim Servo	195	135.75
10707-1	NAV Converter	441	117.75
52D67M	Horizon	T7645A	17.46
Placed Flight Manual	Supplement in POH.		

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Additional Sheets Are Attached



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NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable alrworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

The following Equipment was installed in this Aircraft 14 SEPT 1988 with no Form 337. All of this equipment is shown on Weight And Balance Supplement dated 14 SEPT 1988 Performed conformity check of existing King KN64, serial #16942, DME using King

Installation Manual part #006-00144-0006. The KN64 is connected to the Aircraft electrical buss thru a 3 amp circuit breaker and is located 14.25 inches aft of datum. The KA 60 DME Antenna is located 69.0 inches aft of datum.

Performed conformity check of existing 3M 1000 Weather Mapping System per BF Goodrich Installation Manual part #78-8051-9150-5. The WX1000 Processor, serial #UXP0780050; is connected to the Aircraft electrical buss thru a 4 amp circuit breaker and located 125.7 inches aft of datum. The WX1000 Antenna, serial #UXA07800050, is located 168.0 inches aft of datum. The WX1000 Display, serial #UXD0780050, is located 10.0 inches aft of datum.

Performed conformity check of existing Argus 5000, serial #1425, Moving Map per Argus Installation Manual part #5003. The Argus 5000 is connected to the Aircraft electrical buss thru a 2 amp fuse and is located 10.0 inches aft of datum.

Additional Sheets Are Attached

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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.) INSTALLED BA-6210 FILTER ASSEMBLY IN ACCORDANCE WITH INSTALLATION INSTRUCTIONS BA-6206 DATED SEPTEMBER 12, 1997 OR LATER FAA APPROVED REVISION. NO WEIGHT AND BALANCE CHANGE. END Additional Sheets Are Attached

BRACKETT AIR FILTERS

INSTRUCTIONS

ASSEMBLY PART NO. BA-6210

APPROVED ON AIRCRAFT MODELS: SEE FAA APPROVED APPLICABILITY LIST

FAA APPROVAL DATA: STC-SA71GL List No. 1 FAA-PMA Supplement No. 1

Your new polyurethane air filter element has been designed to give maximum dust collecting efficiency, good air flow, lightweight, and economical replacement. The element has been treated with a special treatment called a wetted agent and is approximately 98% efficient. The special wetted agent is the secret to the efficiency of capturing dust and repelling water. The element has also been treated with a fire retardant. For the above reasons, replace the element each 200 hours of use or every 12 months or when 50% covered with foreign material. DO NOT WASH AND REUSE.

INSTALLATION INSTRUCTIONS

- Step 1. Remove original air filter and discard.
- Step 2. Install filter assembly onto aircraft using the same procedure as original.
- Step 3. Remove element BA-6205 from plastic bag and install into frame assembly.
- Step 4. Fasten grill onto frame with the 2 AN3-23A bolts, making sure center bar on grill is under frame.
- Step 5. To replace future elements remove the grill with the 2 screws only.
- Step 6. NOTE: After initial installation of filter assembly fill out FAA Form 337 for return to service. On future replacement of parts this form will not be required.
- SERVICING: Under normal conditions, replace filter element, PN BA-6205, after 200 hours use or 1 year intervals. Under severely dusty conditions, <u>check daily</u> and replace when element is 50% covered with foreign material.

Instruction Sheet Part No. BA-6206 Date: Rev. 8-29-88 Mfg. by: Brackett Aircraft Co., Inc. Kingman, AZ

URACIMETT ALRORAUT CO., INC. STC SA71GL APPROVED MODEL LIST NO. 1

PAGE NO.15

Maule: M4, M4C M4S, M4T, M-4-22C, M-4-220C, M-4-AIRPLANE MODEL M20D, M20G. M20A, Mooney M20 235C, M-5-180C M-5-210TC, M-6-180S, M-4-180T, M-5-220C, M-5-Cessna 172RG M-4-180C, 220S, M-4-220T, Mooney M20J 80. BRACKETT FILTER MODEL 235 BRACKETT FILTER MODEL M-6-235, m20B, BRACKETT FILTER MODEL, BA-6210 M-4-M20C, T.C. NO. 3A17~ 3A23 273 2A3 BA-5810 BA-6110 later FAA approved revision. 1ation instructions No. BA-5806 dated June 8, 1982 or Install BA-5810 Filter Assem-bly in accordance with instal-DESCRIPTION OF TYPE DESIGN CHANGE bly in accordance with instal-lation instructions No. BA-7105 dated August 10, 1975 Install BA-6110 Filter Assemor later FAA approved revision Install BA-62:0 Filter Assem-bly in accordance with instal dated September 12, later FAA approved revision. lation instructions BA-6206 AIRCEAFT CERGICS I ON SPFICE APPLICABILITY LIST INITIALS: APPLICABILITY LIST FAA APPROVED APPLICABILITY LIST **JUN I 8 1992** 1977 or BA-.-APPROVED DATE July May 5, 1976 Mar. Jan. 20, 22, 1984 29, 1980 1982 ı

> Supplemental Type Certificate Bepartment of Transportation—Kideral Aviation Administration United Status of America

Number SA71GL

This confifcate, is Brackett Aircraft Company, Inc.

entifies that the change in the type design for the following product with the timilations and conditions

Monforateshevifued heronomeds. We aincombined on contract of Barl Specified under certification basis on the applicable Data Sheets or Aircraft Specifications. Degulations.

Chipinal Product - Type Certificato Sumber: Alake: ß shown an Approved Model List(s)

Midel:

Description of Type Dusion Charge: Replace existing engine air filter frame assembly with Brackett Aircraft Specialties frame assembly in accordance with Approved Model List.

NOTE: FAA- Approved Model List(s) form a part of this certificate.

Linviations and Grinditions: This approval should not be extended to other aircraft of this model on which other previously approved modifications are incorporated unless it is determined by the installer that the interrelationship between this change and any of those other previously approved modifications will introduce no adverse effect upon the airworthiness of that aircraft.

This certificate and the supporting data which is the hears for approval shall romain in affect sutilisur

rendered, suspended, rowhed, or a termination date is etherwise established by the Administrator of the

Federal Shriation Staministration

Shal afapphicalim: January 9, 1975

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My direction . Supervisor, Aircraft Modification Section the Staministrator 102

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

FAA FOLM \$110-2 (10-68)

FAA AIRCRAFT REGISTRY CAMERA NO. 2 DATE: 3-19-93

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F.	CERTIFICATION - I hereby certify	ly that I am the registered owner (or h	nis agent) of the aircra	It described above; that the airc	raft is registered with the l	Federal Aviation Administration in
	escribed.	Federal Aviation Act of 1958, and a	pplicable Federal Avi	ation Regulations; and that the	aircraft has been inspec	ted and is airworthy for the flight
04	ATE	NAME AND TITLE (Print or typ)		SIGNATURE	
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х	A. Operating Limitations and Mar 91.31 as Applicable	rkings in Compliance with FAR	I	X G. Statement of Conform	ity, FAA Form 8130-9 (Al	tach when required)
- ¥				H. Foreign Airworthiness	Certification for Import /	Aircraft
-	B. Current Operating Limitations			. (Attach when required	,	•
		ate (Amark when everying d)		L Previous Airpeorthinese	Certificate Issued in Acc	cordance with
	C. Data, Drawings, Photographs,	ecc (Attach when required)				
X	C. Data, Drawings, Photographs, D. Current Weight and Balance Ir			FAR	CAR	(Onginal Attached
X	D. Current Weight and Balance Ir	nformation Available in Aircraft		FAR	CAR	(Original Attached
X	D. Current Weight and Balance Ir	nformation Available in Aircraft AA Form 337 (Attach when required		FAR	CAR	(Original Attached

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PAA AIRCRAFT REGISTRY CAMERA NO. 3 DATE: 11-9-80

STA	NDARD AIR WORTHINESS CERTII	SERIAL 4. CATEGORY
NATIONALITY AND REGISTRATION MARKS	2. MANUFACTURER AND NODEL MODEL MODEL	Normal
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	1. MAKE	ION I ~ AIRCRAFT	
	Mooney Aircraft Corporation	M20J	
	3. SERIAL NO. 24-1658	ALREGISTRATION NO. N5216E	
		TION II - ENGINE	
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	3. SERIAL/NO.049 . tino eros/jerne eest . 1992/14. 1 have - ns to sussi lanight addred reasonings SECTIO		
	1. NAKE		
	3, BLADE MODEL	4. HUB SERIAL NO.	
	5. BLADE SERIAL NOS.		~
		IV - CERTIFICATION	
•	I hereby certify that:	ಕಾರ್ಯ ಶಸ್ತ್ರಿ ಕ್ರಮಿಂಗ್ ಕ್ರಾರ್ ಪ್ರಕಟ್ಟೆ ಕ್ರಾಮಿಂಗ್ ಪ್ರತಿ ಪ್ರತಿ	
	A. I have complied with Section 21.33(a).	n seguri en la seconda en la seconda de l	
	is in a condition for safe operation, and was fligh	certificate only (FAR 21 Subpart F), conforms to its type certificate only (7-1-88	te,
	to at a condition for sale operation, and was might	(Date)	
	C. The engine or propeller described above, presente	d herewith for type certification, conforms to the type design there	or.
	D. The engine or propeller described above produced	under type certificate only (FAR 21 Subpart F), conforms to its typ	e
•	certificate and is in a condition for safe operation	. The engine or, if applicable, the variable pitch propeller was sub	jecte
	by the manufacturer to a final operational check of	t	
	talin se an dregter modifierer in titler og en de en til sin F	n en ante a ser en ante a ser en a ser en altre a s La constante de la constante de	
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A Carlos Martin Mart

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VILLE INSTRUCTIONS

This form should be submitted to a representative of the Admirristrator under the following circumstances:

By the applicant for a type certificate or a supplemental type certificate at the time he presents an aircraft or parts thereof to the FAA for tests.

2. By the applicant for a type certificate or a supplemental type certificate for each engine or propeller submitted for type certification.

3. By the type certificate holder or licensee manufacturing products under a type certificate only, upon the initial transfer by him of the ownership of each product or upon application for the original issue of an aircraft airworthiness certificate, or an Airworthiness Approval Tag (FAA Form 8130-3).

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This form should be completed as follows:

Section I. Aircraft. - Complete the pertinent part of only this section when certification covers an aircraft or part thereof. .20 × 34193 SECTION IV - CERTIFIC: YION

Section II. Engine. Complete this section when certification covers an engine.

Section III. Propeller. Complete this section when certification covers a propeller. and draw boundary

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Section IV. Certification. ະເອັສນະລ**ີນີ້ຫຼາຍວ**່າເຖິງການ<mark>ພ</mark>າຍ ກຳລະ ລາວ ແລະ ແລະ ແມ່ນ Item A. Check this block when an aircraft or part thereof is presented for flight or ground tests during type certification or supplemental type certification.

Item B. Check this block when the holder or licensee of a type certificate only, initially transfers the owner-ship of an aircraft manufactured under that type certificate, or applies for the original issuance of an airworthiness certificate.

Item C. Check this block when an engine or propeller is presented for type certification.

Item D. Check this block when an engine or propeller is presented for airworthiness approval and insert the date the product completed a final operational check.

The certification must be signed by an authorized person who holds a responsible position in the manufacturing organization.